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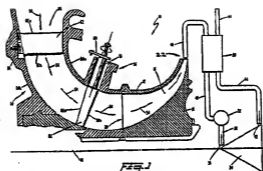
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54 Apparatus and method for controlling mass flow rate in rotary compressors.

57 Two sets of guide vanes (52) and (62) are positioned in the combustion air flow path (54) in the inlet duct (50) of a rotary compressor (10) of a gas turbine engine for controllably varying the air mass flow rate according to turbine load conditions. The upstream set (62) which provides a fixed, initial degree of swirl relative to compressor rotational direction and axis (18) and the controllably moveable downstream set (52) which provides a final degree of swirl cooperate to provide controllable swirl over the range of about 0° to 32° in the inlet air incident upon the compressor blades (16). The two guide vane sets (52) and (62) are separated by a distance sufficient to allow turbulence induced by the first set (62) to fully decay before the second set (52) is encountered.



Apparatus and method for controlling
mass flow rate in rotary compressors

The present invention involves improvements in apparatus and methods for controlling mass flow rate in rotary compressors, especially but not exclusively in compressors used in recuperated gas turbine engine applications.

The power output of conventional gas turbine engines can be varied by changing the turbine inlet temperatures, such as by reducing fuel flow. However, it is known that substantial increases in the part load efficiency can be achieved if the mass flow rate of the combustion air is reduced to maintain high turbine inlet temperatures, particularly in recuperated gas turbine engine applications. It is also known that precise control of the combustion temperature can lead to reductions in the amount of undesirable hydrocarbon and nitrogen oxide emissions.

Previous attempts to change or control the air mass flow rate in externally driven compressors, such as compressors driven by an electric motor or compressor section of gas turbine electrical power generator units, involve the use of guide vanes in the compressor inlet that are moveable to induce swirl in the incoming air to change the angle at which the inlet air enters the compressor blades. Most compressors are designed to have blade shapes and angles of attack orientation selected to obtain optimum mass flow rate at rated speed. These design point conditions generally presuppose inlet air incident at a fixed, predetermined angle relative to the axis of rotation of the compressor. Inlet air flow incident at angles different from the design value, such as occurs when swirl is introduced or the amount of swirl is changed, causes the mass flow rate through the apparatus to change from the design value.

A problem with conventional swirl-inducing guide vane apparatus used to vary the mass flow rate in compressors is

that the maximum degree of turning or swirl achievable without substantial separation with a single set of vanes is approximately 15°, while variations in the swirl angle of about 30° may be desirable in certain applications, such as compressors used in gas turbine engines, in order to achieve high thermal efficiency throughout the entire operating range. Although moveable vanes with turning angles greater than 15° have been attempted, these are susceptible to severe separation and consequent losses. Various attempts have been made to circumvent this problem such as by the use of two-piece articulated vanes having a fixed leading portion and a moveable trailing or tail portion. Another proposed solution utilizes two sets of vanes, a fixed set immediately upstream of a moveable set to achieve essentially the same function as the articulated vanes. These solutions are not satisfactory as less than the desired range of turning can be achieved in practice commensurate with the requirement for a reasonably low aerodynamic loss.

It is believed that the deficiencies of the prior art stem in large part from the close aerodynamic coupling between the front and rear portions of the articulated vanes and between the upstream and downstream separate sets of guide vanes. While it is believed that the reason for the deliberately close coupling in the prior art was to take advantage of the known flow profile incident upon the downstream vane portion or vane set, the lack of smooth, undisturbed streamline flow on the downstream elements can result in unwanted turbulence and premature boundary layer separation leading to high losses in turning conditions. The losses imposed on the air flow could also be substantial in the non-turning conditions because of the proximity of the fixed vanes to the compressor blades. The improvements of the present invention are intended to circumvent these problems.

In accordance with the purpose of the invention, as embodied and broadly described herein, the improvement in

rotary apparatus for compressing a compressible fluid of the kind having a plurality of compressor blades mounted on a rotating hub positioned in a compressor housing, a preferred fluid flow path extending through the housing, the housing
5 having a duct portion extending upstream of the compressor blades relative to the fluid flow path and determining, in part, the fluid flow path, comprises means for controllably varying the fluid mass flow rate through the compressor including a first set of guide vanes in the said duct
10 portion for imparting an initial degree of swirl to the fluid entering the duct portion relative to the direction of rotation of the compressor hub; and a second set of guide vanes positioned in the said duct portion upstream of the compressor blades, the vanes of the second set being
15 moveable about their axes, the second vane set being positioned a distance downstream of the first vane set along the fluid flow path sufficient to permit substantial decay of the turbulence imparted to the flowing fluid by the first vane set prior to the swirling fluid reaching the second,
20 moveable vane set, the second moveable vane set serving for changing the degree of swirl in the fluid to a final degree of swirl corresponding to a desired compressor fluid mass flow rate.

Preferably, the first vane set is fixed, and the
25 individual guide vanes in the first vane set are configured and oriented to impart about $+10^\circ$ to $+15^\circ$ of swirl to the incoming fluid relative to the axis and direction of rotation of the compressor.

It is further preferred that the first vane set and the
30 second moveable vane set cooperate to provide final fluid swirl of from about 0° to $+32^\circ$ at the inlet to the compressor blades, relative to the axis and direction of rotation of the compressor.

Also in accordance with the present invention, as
35 embodied and broadly described herein, the method of throttling the compressible fluid mass flow rate through a

rotary compressor of the kind having a plurality of compressor blades on a rotating hub and having an inlet region including a duct determining, in part, the flow path of the incoming fluid to the compressor blades, comprises the steps of imparting a first, initial degree of swirl to the incoming fluid relative to the axis and direction of rotation of the compressor; removing turbulence in the swirling fluid induced by the initial swirl imparting step; and imparting a second, controllably variable degree of swirl to the swirling fluid to change the degree of swirl in the fluid to a desired value prior to admitting the fluid to the compressor blades, the aforementioned steps being accomplished in the compressor inlet region.

The accompanying drawings illustrate one embodiment of the invention and, together with the description, serve to explain the principles of the present invention. Reference will now therefore be made in detail to the present preferred embodiment of the invention, an example of which is illustrated in the accompanying drawings, in which:-

Fig. 1 is a schematic partly radially sectional view of part of a compressor apparatus associated with a single shaft gas turbine engine and embodying the present invention;

Fig. 2a is a diagram showing the inlet air flow path through the compressor section for low power operation of the gas turbine engine shown in Fig. 1; and

Fig. 2b is a diagram showing the inlet air flow path through the compressor of Fig. 1 at high turbine power operation.

In Fig. 1, there is shown schematically a single shaft compressor 10 used to increase the pressure of a gaseous fluid such as air. The compressor 10 includes a housing 12 surrounding a hub 14 on which are mounted compressor blades 16. The hub 14 is rotatable about an axis 18 (represented by a chain-dotted line in the Fig. 1). The compressor 10 is shown in Fig. 1 to be of the centrifugal type, with an

entrance 20 to the compressor blades 16 predominantly in the axial direction in relation to the axis 18 and with the compressor gas leaving the compressor 10 at an exit 22 substantially in the radial direction. However, the improvements of the present invention are not restricted to use with centrifugal compressors, and the scope of the present invention includes axial compressors as well as mixed axial and radial flow compressor devices.

The improvements constituting the present invention enable the pressure ratio and the mass flow rate to the compressor 10 to be controlled essentially independently of the rotational speed of the compressor. This is an especially important advantage in certain applications such as where the compressor is driven at essentially constant speed such as by a synchronous device or where, such as shown in Fig. 1, the compressor is used in a recuperated single shaft gas turbine engine application. The present invention also can be utilized to advantage in a two shaft machine because although some decrease in the gas mass flow rate occurs with the decrease in rotational speed of the gas generator in such machines, additional reductions can be achieved using the present invention.

Fig. 1 shows the compressor 10 associated with a heat exchanger 20, a combustor 32 and a turbine 34, all illustrated schematically. According to well-known principles, the compressed air emanating from the compressor exit 22 is channeled by appropriate ducting 40 through the heat exchanger 30 where it is heated prior to admission to the combustor 32, such as by the exhaust gases channeled from an exit 42 of the turbine 34 by ducting 44. The heated compressed air is then combusted with fuel in the combustor 32 and the combustion gases are conveyed by ducting 36 and admitted to the turbine 34 at a turbine inlet 38 for subsequent expansion and extraction of mechanical work. The turbine 34 is shown co-axial with the axis 18 of the compressor 10, but other configurations can be implemented

depending upon the particular application.

It is understood that the aforementioned heat exchanger 30, and ducting 36, 40, 44 may be constructed as integral parts of housing 12 and/or the turbine housing (not shown) and are depicted as separate components in Fig. 1 merely for convenience of description. Also, the preheating of the compressed air can be accomplished by channeling the compressed air emanating from the compressor exit 22 past various structural components such as the combustor 32 housing (not shown) and the turbine housing for cooling these components. Although such an arrangement may not have all the thermodynamic advantages of a fully recuperated gas turbine engine, other advantages, such as extended lifetimes for structural and rotating components and better control over hydrocarbon and nitrogen oxide (NOX) emissions, may be achieved. For example, it is desirable to utilize lean burning combustors to control NOX emissions but this requires precisely controlling the fuel/air ratio over the entire operating range. It is far easier to control the fuel/air ratio if the air mass flow rate can be controlled during part load operation, such as by use of the present invention. Thus, the scope of the present invention is intended to include these latter applications as well as the recuperated gas turbine engine application depicted in Fig. 1.

In accordance with the present invention, means are provided for controllably varying the fluid mass flow rate through the compressor, the means including a set of fixed, moveable guide vanes positioned upstream of the compressor blades. As embodied herein, the compressor housing 12 has an inlet duct portion 50 extending upstream of the compressor blades 16 along the inlet air flow path through the compressor 10 (designated by arrows 54). The inlet duct portion 50 is shown as having a shroud side 50a, a hub side 50b, and an air inlet 66. A set of guide vanes 52 having leading edges 52a and trailing edges 52b are positioned in

the compressor inlet duct portion 50 upstream of the compressor blades 16, and the function of the vanes 52 is to impart a final controllably variable degree of swirl to the fluid in the flow path 54, relative to the direction of the rotation of the hub 14 about compressor axis 18, the final degree of swirl corresponding to the desired compressor mass flow rate. The vanes 52 are attached to the inlet duct portion 50 by a blade mounting assembly 56 which provides for rotational movement of the vanes 52 about the respective vane longitudinal axis 58 such as to present a varying angle of attack to the incident air flow 54. Changes in the angular orientation of the vanes 52 about the axis 58 are accomplished through a lever assembly 60 attached to the vanes 52, but any suitable mechanical, hydraulic or other actuating mechanism can be used. Actuating mechanisms such as the lever assembly 60 can be automatically controlled by conventional controller means (not shown) in accordance with desired operating conditions, or would be immediately evident to those skilled in the art.

Preferably, the vanes 52 are positioned in the inlet duct portion 50 adjacent the compressor entrance 20 and proximate the compressor blades 16 along the flow path 54 for the following reasons. The vanes 52, however, should be spaced from the compressor blades 16 by a distance sufficient to allow any wake generated in the inlet air by the vanes 52 to close before the inlet air reaches the blades 16. The final flow profile incident upon the compressor blades 16 will be determined by the vanes 52 as will be discussed hereinafter, and the proximity of the vanes 52 to the compressor blades 16 will secure definition of the final swirl profile insofar as there will be minimal interaction with the inlet duct portion 50. It is also preferred for essentially the same reasons that the vanes 52 be located in a region of the duct portion 50 wherein the average streamline velocity at the leading edge of the vanes is at least about 70% and, more preferably, more than about

80% of the average streamline velocity at the compressor entrance 20.

In accordance with the present invention, there is further provided a set of initial guide vanes 62 for
5 imparting an initial degree of swirl to the inlet air entering the compressor, relative to the axis 18 and direction of rotation of the compressor, thus dividing or sharing the total turning between the two separate sets of guide vanes 62 and 52. As embodied herein, the vanes 62 are
10 positioned in the inlet duct portion 50 upstream of the moveable vanes 52 along the flow path 54 and near an air inlet region 66. Thus, the inlet air entering the compressor 10 along the flow path 54 at the inlet 66 is influenced first by the guide vanes 62 and then second by
15 the guide vanes 52 before being admitted to the compressor blades 16.

Preferably, the vanes 62 present to the air flow entering the inlet duct portion 50 at the inlet 66 along the flow path 54 an angle of attack that remains constant in
20 time during operation of the compressor 10 over the entire range of load conditions, although the angle may vary spatially along the vane axis to achieve a desired aerodynamic flow pattern in a particular inlet duct configuration. For instance, in the configuration for duct
25 portion 50 shown in Fig. 1, the velocities near the shroud side 50a will be higher than the inlet air velocities near the hub side 50h. Thus, the portion of the vane 62 near the hub side may have a greater angle of attack than the shroud side portion giving rise to a "twist" in the profile of the
30 vanes 62. The vanes 52 can also have a twist to further match the incident flow profile, that is, to provide a spatially constant angle of attack to the air flow incident from the vanes 62. In general, the "twist" in the vanes 52 will be less than that in the vanes 62.

35 To achieve an angle of attack that is constant in time, the vanes 62 can be permanently fixed in the inlet duct

portion 50 such as by making them an integral housing structural member or can be attached by suitable fastening means. As shown in Fig. 1, the vanes 62 are fastened by a bolting mechanism 64 to permit adjustment changes in the angle of attack of the vanes 62 during initial assembly of the gas turbine apparatus or during subsequent servicing outages, in order to achieve optimum results.

Importantly, the distance between the initial guide vanes 62 which impart a constant degree of initial swirl to the inlet air and the final, moveable guide vanes 52 which impart a final degree of swirl to the inlet air, depending upon operating load condition, is such as to permit the turbulence induced by the vanes 62 to become essentially decayed in order to provide substantially invariant streamline flow across the flow area of the flow path 54 immediately upstream of the vanes 52. This positioning enables the moveable vanes 52 to be aerodynamically decoupled from the guide vanes 62 to the extent that premature boundary layer separation on the vanes 52 will not be induced by asymmetrical, undecayed wake from the vanes 62.

It is estimated that spacing the trailing edges 62b of the vanes 62 from the leading edges 52a of the vanes 52 by a distance of greater than or equal to about one chord length of the vanes 62 should provide essentially fully decayed turbulent flow incident upon the vanes 52. Preferably, the vanes 62 should be located in a region of relatively low air flow velocities so that the full load losses are small and on the order of about less than 30% of the average streamline flow velocity at the compressor entrance 20 and, more preferably, less than about 10%. For reasonable duct configurations, such as that shown in Fig. 1, this requirement entails a physical separation distance of at least one chord length of the vane 62 and usually two to three chord lengths, thus ensuring fully decayed flow at the leading edges 52a.

In the centrifugal apparatus disclosed in Fig. 1, the vanes 62 are located in the inlet region 66 of the inlet duct portion 50. The flow path 54 in the inlet region 66 is predominantly in the radial direction. The moveable guide
5 vanes 52 are located at the exit of the inlet duct portion 50 near the compressor entrance 20 in a region where the flow path 54 is predominantly axial and where average streamline flow velocities on the order of about 250
meters/second occur. One skilled in the art could determine
10 without undue analysis or experimentation an appropriate separation distance for a given compressor inlet duct configuration given the present disclosure.

It is also preferred that the guide vanes 62 are configured and oriented in the inlet duct portion 50 to
15 impart about +10° to +15° of initial swirl to the incoming fluid under all turbine load conditions, the degree of initial swirl being measured immediately upstream of the vanes 52 along the flow path 54, relative to the axis 18, and with the direction of rotation of the hub 14
20 establishing the positive direction. With reference to Figs. 2A and 2B, the angle of initial swirl is shown by the angle α which is positive as it is in the direction of rotation of the hub 14 (designated by arrows).

It is still further preferred that the moveable vanes
25 52 are configured and oriented by a lever assembly such as the assembly 60 to impart a further final degree of swirl in the fluid incident upon the compressor blades 16 (represented by the angle β in Figs. 2A and 2B) ranging from about 0° to +32° for the maximum and minimum turbine load
30 conditions, respectively. For the apparatus in Fig. 1, the guide vanes 52 should be capable of changing the relative direction of the air incident on the vanes 52 from the vanes 62 from about -20° to +20° depending upon the turbine load.

With specific reference to Figs. 2A, 2B which depict
35 the invention being used in the compressor component of a gas turbine engine operating at minimum and maximum load

conditions, respectively, it is clear that the orientation of the vanes 52 at minimum turbine load is such as to increase the positive degree of swirl induced by the vanes 62 while at maximum load conditions the orientation of the moveable vanes 52 is such as to impart negative swirl to eliminate or correct the final degree of swirl in the combustion air to the 0° maximum load design condition for the compressor 10 shown in Fig. 1. If the particular turbine apparatus includes a compressor having a design point with a finite, non-zero (positive or negative) degree of swirl, the preferred ranges of the settings for initial set of the vanes 62 and final, moveable vanes 52 would be adjusted accordingly, as would be evident to one skilled in the art upon reading this disclosure.

Because of the aerodynamically decoupled nature of the relationship between the vanes 62 and the moveable vanes 52, a certain amount of analysis and/or experimentation will be necessary in a particular application to determine the angular settings and twists of the vanes 62 and the vanes 52 needed to effect the desired final degree of swirl at the entrance to the compressor blades 16, particularly in applications such as that shown in Fig. 1 wherein the compressor inlet duct portion 50 contains substantial changes in the flow path 54 direction between the vanes 62 and 52. However, the penalties in terms of increased design cost are outweighed by the expected increase in overall efficiency of the turbine unit, particularly at low load conditions where large final swirl angles can be achieved on stable basis. This increased low load efficiency also is expected to outweigh the slight degradation in performance at the maximum or design load due to the additional pressure drops incurred in the inlet duct portion 50 to, in effect, first induce swirl and then remove the swirl in the inlet air incident upon compressor 16. Although the use of moveable vanes for the vanes 62 would overcome the deficiency, and is considered to be within the scope of the

present invention, the extra complexity makes such an embodiment not as preferable as the fixed vane 62 embodiment shown and described hereinbefore.

5 It will be apparent to those skilled in the art that various modifications and variations could be made in the improved means for controlling the mass flow rate in a compressor and the corresponding improved method for achieving control of the mass flow rate without departing from the scope or spirit of the invention.

CLAIMS

1. Rotary apparatus for compressing a compressible fluid of the kind having a plurality of compressor blades (16) mounted on a rotating hub (14) positioned in a compressor housing (12,50), a preferred fluid flow path (54) extending through the housing (12,50), the housing (12,50) having a duct portion (50) extending upstream of the compressor blades (16) relative to the fluid flow path (54) and determining, in part, the fluid flow path (54), and

means (52,62) for controllably varying the fluid mass flow rate through the compressor, characterised in that the said means includes:-

a) a first set of guide vanes (62) in the said duct portion (50) for imparting an initial degree of swirl to the fluid entering the duct portion (50) relative to the axis (18) and direction of rotation of the compressor hub (14); and

b) a second set of guide vanes (52) positioned in the duct portion (50) upstream of the compressor blades (16), the vanes (52) of the second set being moveable about their axes, the second vane set (52) being positioned a distance downstream of the first vane set (62) along the fluid flow path (54) sufficient to permit substantial decay of the turbulence imparted to the flowing fluid by the first vane set (62) prior to the swirling fluid reaching the second, moveable vane set (52), the second, moveable vane set (52) serving for changing the degree of swirl in the fluid to a final degree of swirl incident upon the compressor blades (16) corresponding to a desired compressor fluid mass flow rate.

2. Rotary apparatus according to claim 1, wherein the second, moveable vane set (52) is positioned upstream of and adjacent to the compressor blades (16).

3. Rotary apparatus according to claim 1, wherein the duct portion (50) has a predominantly radial inlet portion (66)

and a predominantly axial exit portion (20), the fluid flow path (54) undergoing a substantial angular change in direction through the duct portion (50), and wherein the first vane set (62) is located in the inlet portion (66) and this second moveable vane set (52) is located in the exit portion (20).

4. Rotary apparatus according to claim 1, wherein the individual guide vanes in the first vane set (62) are configured to impart about $+10^\circ$ to $+15^\circ$ of swirl to the incoming fluid relative to the axis (18) and direction of rotation of the compressor hub (14).

5. Rotary apparatus according to claim 1, wherein the individual guide vanes in the second, moveable vane set (52) are moveable to change direction in the fluid incident thereupon by about -20° to $+20^\circ$, the direction of rotation of the compressor hub (14) establishing the positive direction.

6. Rotary apparatus according to claim 1 or claim 4, wherein the first vane set (62) and the second, moveable vane set (52) cooperate to provide fluid swirl of from about 0° to $+32^\circ$ at the inlet to the compressor blades (16), relative to the axis (18) and direction of rotation of the compressor hub (14).

7. Rotary apparatus according to claim 1, wherein the first vane set (62) is essentially fixed.

8. Rotary apparatus according to claim 1, wherein the first vane set (62) is positioned in a region of the duct portion (50) having average streamline fluid velocities of less than about 30% of the average streamline velocities at the compressor blades (16).

9. Rotary apparatus according to claim 8, wherein the first vane set (62) is positioned in a region of the duct portion (50) having average streamline fluid velocities less than about 10% of the average streamline velocities at the compressor blades (16).

10. Rotary apparatus according to claims 1 or 8, wherein

the second, moveable vane set (52) is positioned in a region of the duct portion (50) having average streamline fluid velocities of at least 70% of the average streamline fluid velocities at the compressor blades (16).

11. Rotary apparatus according to claim 9 or 10, wherein the second, moveable vane set (52) is positioned in a region of the duct portion (50) having average streamline fluid velocities of greater than about 80% of the average streamline fluid velocities at the compressor blades (16).

12. Rotary apparatus according to claim 1, wherein at least about one first vane chord length separates the trailing edges of the first vane set (62) from the leading edges of the second, moveable vane set (52).

13. Rotary apparatus according to claim 12, wherein the separation distance is about 2-3 chord lengths.

14. An air-breathing, variable load recuperated gas turbine engine of the kind having a rotary compressor section (10) for compressing the air for combustion, a combustor (32) for combusting the compressed air with fuel to produce

combustion gases, a turbine (34) for recovering mechanical work from the combustion gases, and apparatus (30) for recovering heat values from the combustion gases and heating the compressed air prior to combustion using the recovered heat values, the compressor (10) further including a

plurality of compressor blades (16) mounted on a rotating hub (14) positioned in a compressor housing (12,50), the compressor housing (12,50) having a preferred air flow path (54) extending therethrough, and also having a duct portion (50) extending upstream of the compressor blades (16)

relative to the air flow path (54), the duct portion (50) determining, in part, the air flow path, and

means (62,52) for controllably varying the air mass flow rate through the compressor (10) and to the combustor (32) to maintain highest possible turbine section inlet temperatures over varying turbine load conditions, characterized in that the air mass flow rate varying means

- 16 -

includes

a) a first set of guide vanes (62) in the duct portion (50) for imparting an initial degree of swirl to the fluid entering the compressor housing inlet (66) relative to the axis (18) and direction of rotation of the compressor hub (14); and

b) a second set of guide vanes (52) positioned in the duct portion (50) upstream of the compressor blades (16), the vanes (52) of the second set being moveable about their axes (58), the second vane set (54) being positioned a distance downstream of the first vane set (62) along the fluid flow path (54) sufficient to permit substantial decay of the turbulence imparted to the flowing fluid by the first vane set (62) prior to the swirling fluid reaching said second, moveable vane set (52),

the second, moveable vane set (52) changing the degree of swirl in the fluid to a final degree of swirl corresponding to a predetermined turbine load condition.

15. A method of throttling the compressible fluid mass flow rate through a rotary compressor (10) of the kind having a plurality of compressor blades (16) on a rotating hub (14) and having an inlet region including a duct (50) determining, in part, the flow path of the incoming fluid to the compressor blades (16), the method being characterised by the steps of:-

a) imparting a first, initial degree of swirl to the incoming fluid relative to the axis (18) and direction of rotation of the compressor;

b) removing turbulence in the swirling fluid induced by the initial swirl imparting step; and

c) imparting a second, controllably variable degree of swirl to the swirling fluid to change the degree of swirl in the fluid to a desired final value prior to admitting the fluid to the compressor blades, steps a)-c) being accomplished in the duct (50).

16. A method according to claim 15, wherein the first swirl

imparting step is performed at a location along the fluid flow path (54) a distance upstream of the location where the second variable swirl imparting step is performed sufficient to allow the induced turbulence to decay naturally.

17. A method according to claim 15 or 16, wherein the second, variable swirl imparting step is accomplished by variable angle guide vane means (52) positioned in the fluid flow path (54), and wherein the first swirl imparting step is accomplished by fixed angle guide vane means (62)

positioned in the duct (50) upstream of the variable guide vane means (52).

18. A method according to claim 15, wherein about $+10^\circ$ to $+15^\circ$ of swirl relative to the axis (18) and direction of rotation are imparted to the incoming fluid in the first swirl imparting step, measured immediately upstream of the location in the fluid flow path where the second, variable swirl imparting step is carried out.

19. A method according to claim 15, wherein the second, variable swirl imparting step changes the direction of the incident fluid from about -20° to $+20^\circ$.

20. A method according to claim 15, wherein the second, variable swirl imparting step together with the first swirl imparting step provide a final swirl of between about 0° to $+32^\circ$ prior to admitting the fluid to the compressor

blades (16).

21. A method according to claim 15, wherein the second, variable swirl imparting step is carried out in a region of average streamline fluid velocities of at least about 70% of the average streamline fluid velocities at the compressor blades (16).

22. A method according to claim 15, wherein the first swirl imparting step is carried out in a region of average streamline fluid velocities of less than about 30% of the average streamline flow velocities of the compressor blades (16).

23. A method according to claim 15, wherein the duct (50)

has a predominately radial inlet portion (66) and a predominately axial exit portion (20), the first swirl imparting step being carried out in the inlet portion (66) and the second, variable swirl imparting step being carried out in the exit portion (20).

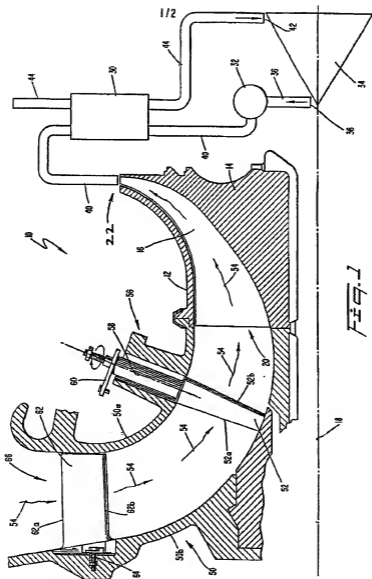
24. A method according to claim 17, wherein the fixed guide vane means (62) includes fixed guide vanes, and wherein the first swirl imparting step location is separated by a distance greater than about one fixed guide vane chord length from the second swirl imparting step location.

25. A method of throttling inlet air flow rate to the combustor (32) in a recuperated gas turbine engine of the type having a rotary compressor (10) with a plurality of compressor blades (16) on a rotating hub (14) and with an inlet region including a duct (50) determining, in part, the flow path (54) of the incoming air to the compressor blades (16), for maintaining highest possible turbine inlet temperatures over variable load conditions, the method being characterized by the steps of:

a) imparting a first, initial degree of swirl to the incoming air relative to the axis (18) and direction of rotation of the compressor (10);

b) removing turbulence in the incoming air induced by the preliminary swirl imparting step; and

c) imparting a second, variable degree of swirl to the incoming air to change the degree of swirl in the air to a final desired value corresponding to a predetermined turbine load condition prior to admitting the air to the compressor blades (16), the steps a)-c) being accomplished in the duct (50).



2/2

